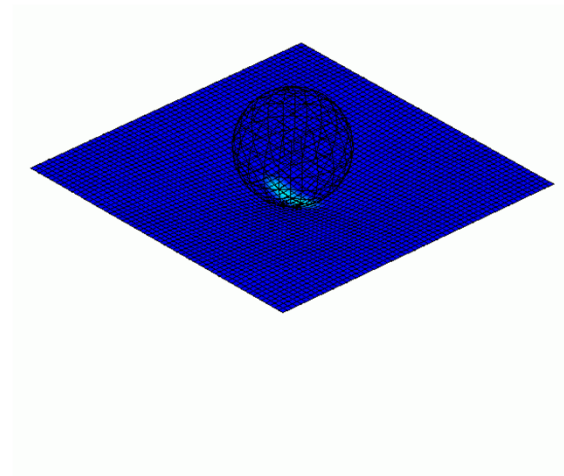




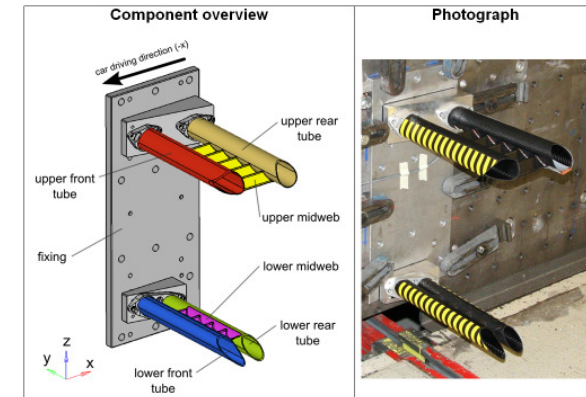
Materialmodellierung und Auslegung von Composite Strukturen für die Crash- und Impactsimulation



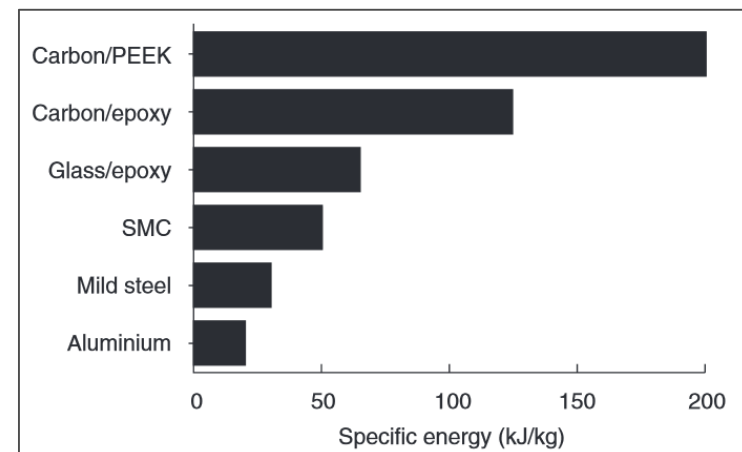
Agenda



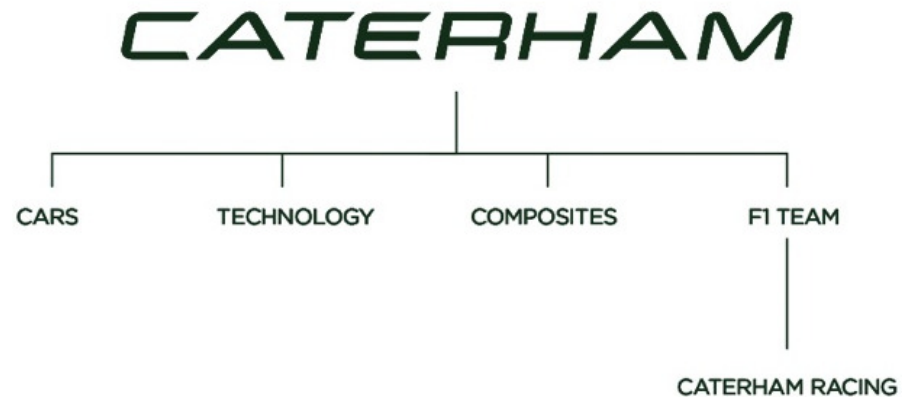
- **Materialmodellierung**
 - **Validierung einer Crashberechnung**
- Auslegung von Composite Strukturen
 - Beschnitt: Optimierung Size/Shape/Shuffling
 - Fertigung: Drapiersimulation



$$SEA = \frac{E_{absorbiert}}{m_{ben\ddot{o}tigt}}$$



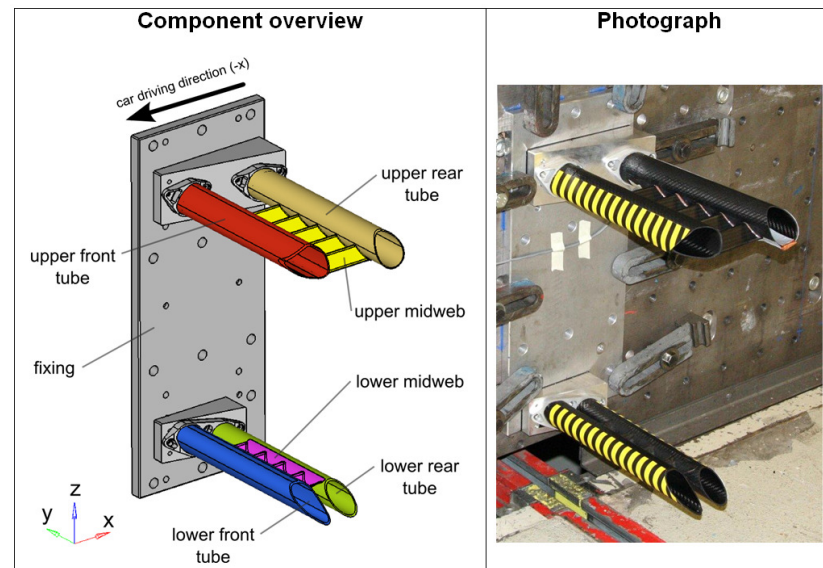
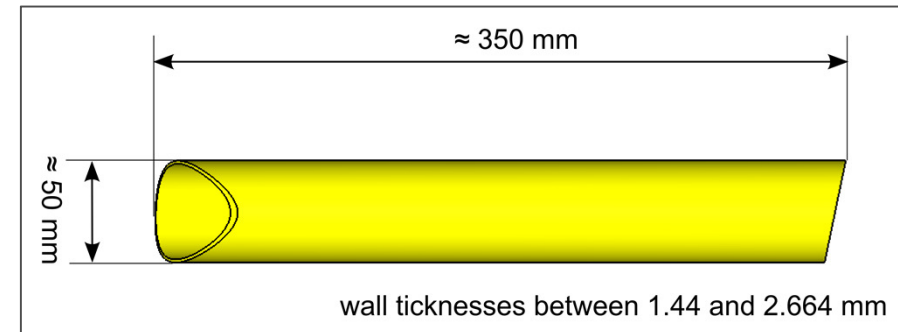
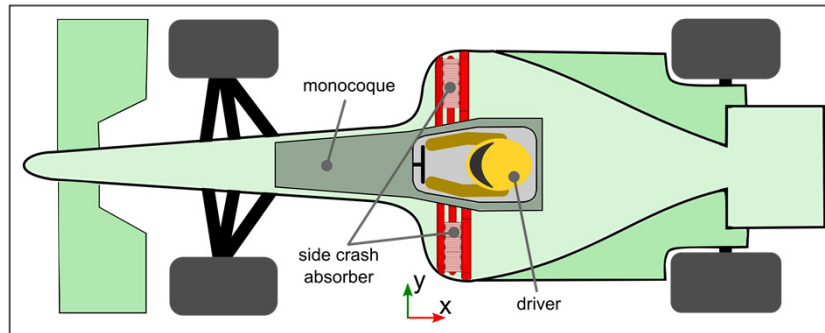
Partner Caterham Composites



Lotus T127 Formel 1
Simulation dynamischer Seitencrash

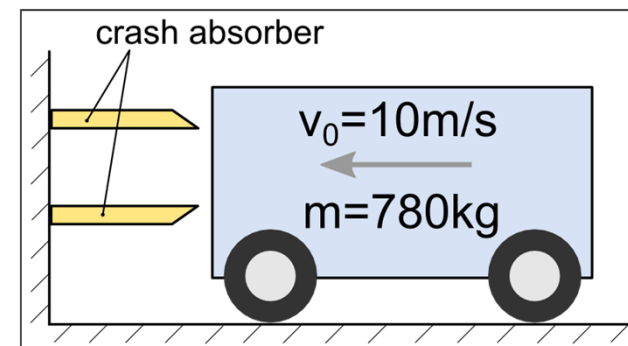
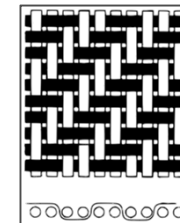
Gegenstand der Untersuchung

Seitencrash-Absorber aus Kohlenstofffaser-Kunststoff-Verbund



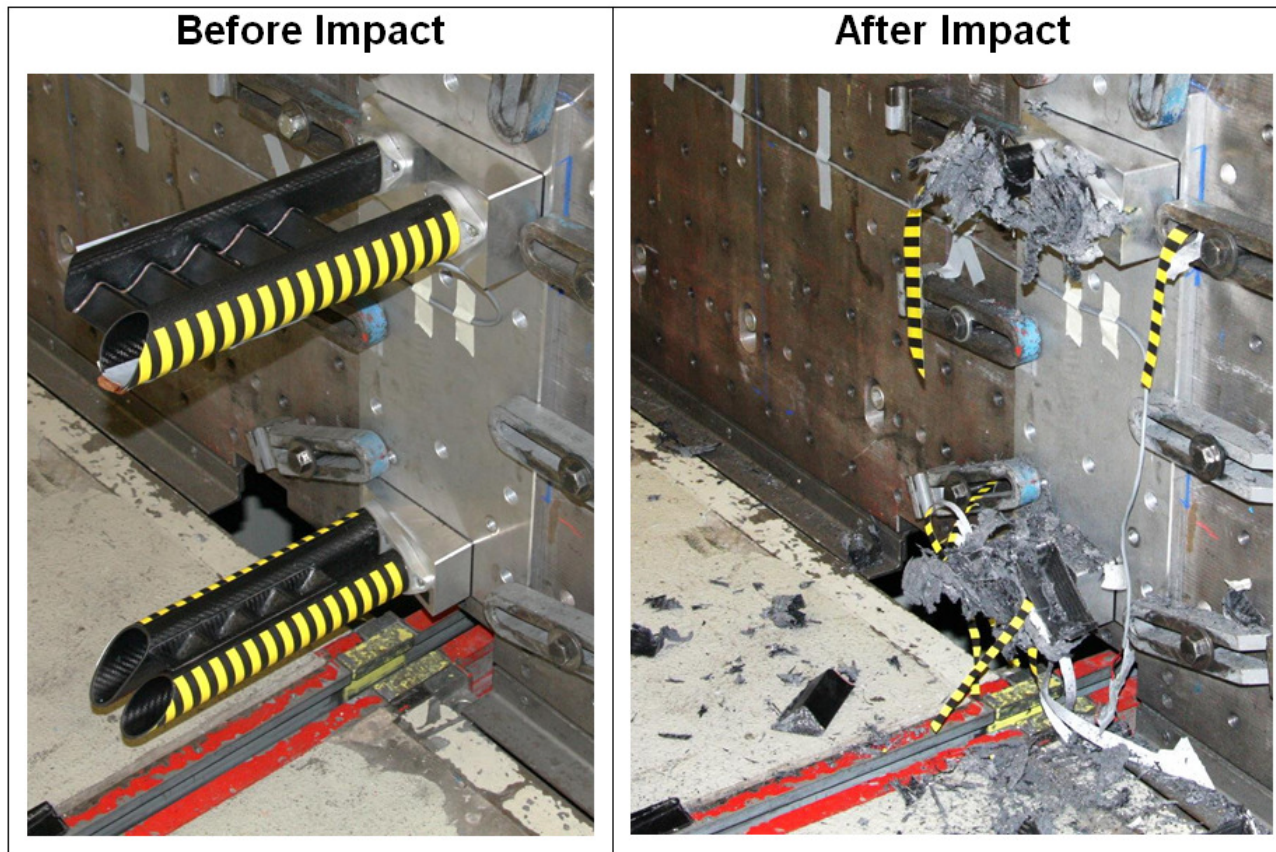
Laminatmaterial

- 2x2 Twill Gewebe



Gegenstand der Untersuchung

Realversuch Seitencrash: Vorher-Nachher-Vergleich

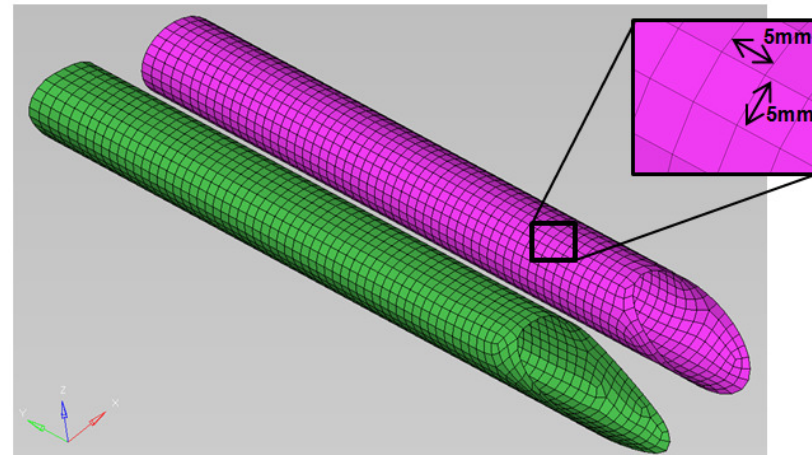
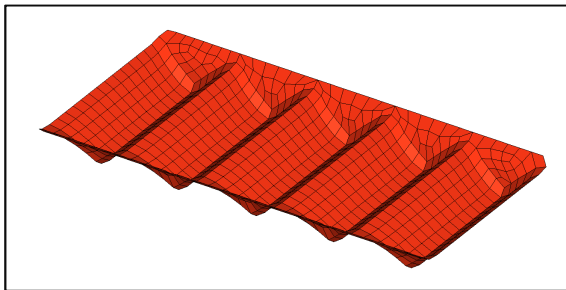
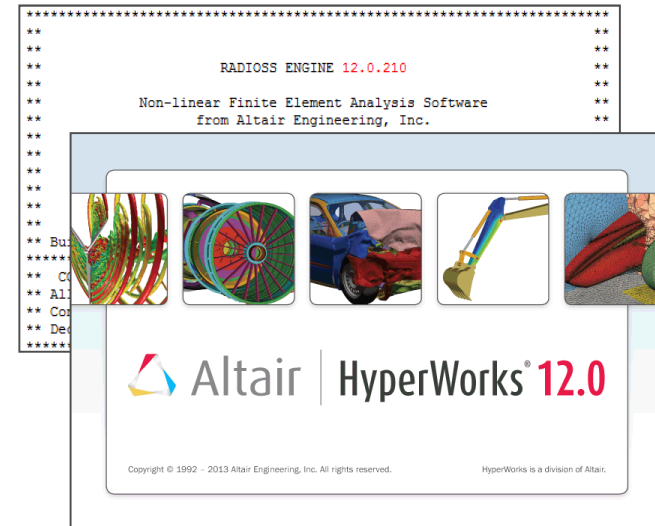


Modellaufbau

Solver: RADIOSS V12.0.210

Vernetzung der Geometrie:

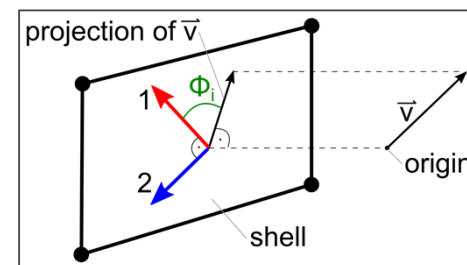
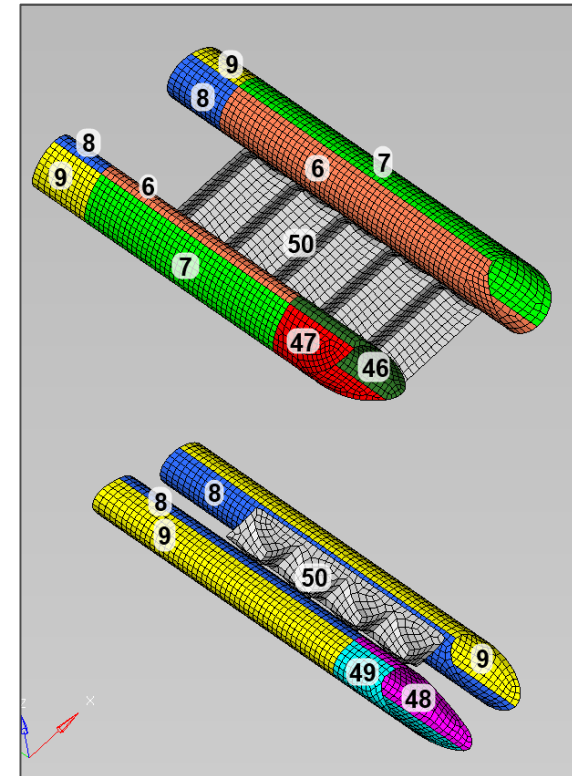
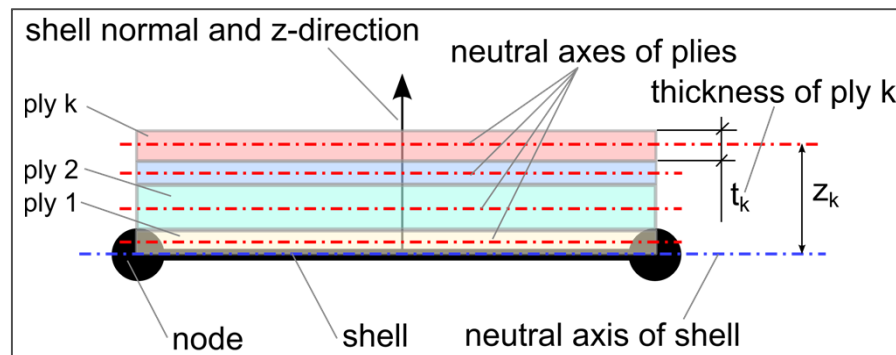
- Grundlage: CAD-Model
- 5x5mm 4-Knoten-Schalenelemente (quads)
- eine Schale pro Laminatdicke
- Anzahl: insgesamt ca. 7500 Elemente



Modellaufbau

Elementeigenschaften (*properties*):

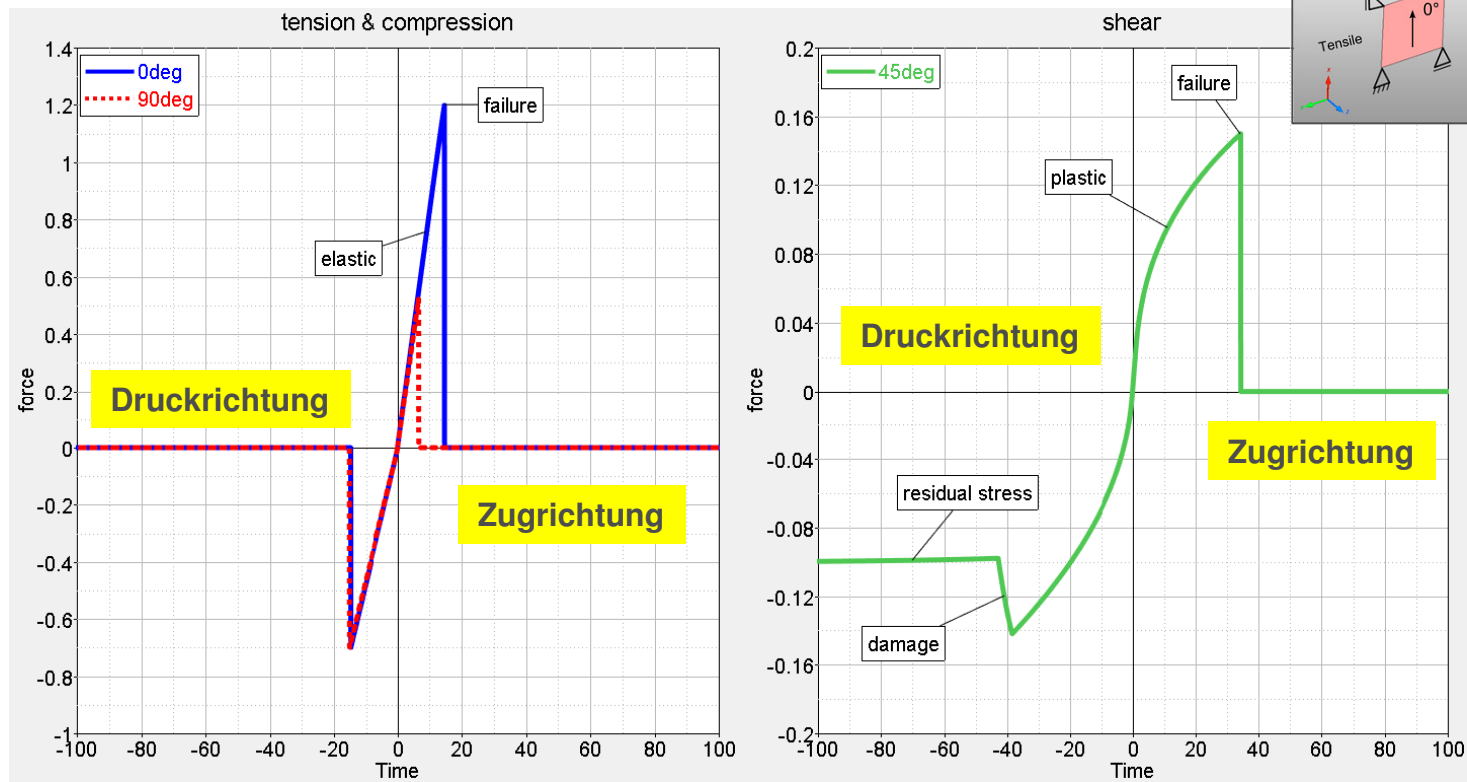
- Definition des Laminataufbaus durch den Sandwich Schalen Ansatz
- Jede *property* erhält Informationen zu **Anzahl** der Lagen, sowie ihre jeweilige **Position**, **Dicke**, **Orientierung** und **Material**



Modellaufbau

Materialmodell:

Elasto-plastisch-orthotrop (exemplarisch)



Radioss Composite Material Model



```

#---1---|---2---|---3---|---4---|---5---|---6---|---7---|---8---|---9---|---10---|
/MAT/COMPSH/10
Altair composite material
#      Init. dens.      Ref. dens.      Linear
#      1.5000E-6      0
#      E11      E22      NU12      IFLAG      E33
#      42      40      .05      1      0.50
#      G12      G23      G31      EPSF1      EPSF2
#      3.4      3      3      0      0
#      ESPT1      EPSM1      EPST2      EPSM2      Dmax
#      0      0      0      0      .9999
#      Wpmax      Wpref      Ioff
#      0      0      5
#      C      EPS      ALPHA      Nonlinear      Icc
#      0      0      0      0
#      sig_trac_1      B_1T      N_1T      SIGMA_1MAXT      C_1T
#      0.1      25      .10      0      0
#      EPS_1T1      EPS_2T1      SIGMA_RST1      Wpmax_trac_1
#      0      0      0      0
#      sig_trac_2      B_2T      N_2T      SIGMA_2MAXT      C_2T
#      0.1      20      .10      0      0
#      EPS_1T2      EPS_2T2      SIGMA_RST2      Wpmax_trac_2
#      0      0      0      0
#      sig_comp_1      B_1C      N_1C      SIGMA_1MAXC      C_1C
#      .0050      2000      .5      0      0
#      EPS_1C1      EPS_2C1      SIGMA_RSC1      Wpmax_comp_1
#      0      0      0      0
#      sig_comp_2      B_2C      N_2C      SIGMA_2MAXC      C_2C
#      .0050      2000      .5      0      0
#      EPS_1C2      EPS_2C2      SIGMA_RSC2      Wpmax_comp_2
#      0      0      0      0
#      sig_12      B_12T      N_12T      SIGMA_12MAXT      C_12T
#      .0040      83.0      .31      0      0
#      EPS_1T12      EPS_2T12      SIGMA_RST12      Wpmax_trac_12
#      0.075      0.085      0.05      0
#      GAMMA_INI      GAMMA_MAX      Dmax
#      1E31      1E31      .9999
#      Fsmooth      Fcut
#      0      0
#---1---|---2---|---3---|---4---|---5---|---6---|---7---|---8---|---9---|---10---|

```

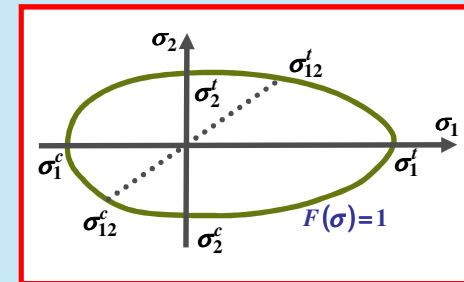
Composite Material Model: Yield Stress



```

#---1---|---2---|---3---|---4---|---5---|---6---|---7---|---8---|---9---|---10---|
/MAT/COMPSH/10
Altair composite material
#      Init. dens.      Ref. dens.
#      1.5000E-6        0
#      E11              E22              NU12      IFLAG
#      42              40              .05        1
#      G12              G23              G31        EPSF1
#      3.4              3              3          0
#      ESPT1            EPSM1            EPST2      EPSM2
#      0                0              0          0
#      Wpmax            Wpref            Ioff
#      0                0              5
#      C                EPS              ALPHA      Icc
#      0                0              0          0
#      sig_trac_1        B_1T            N_1T        SIGMA_1MAXT      C_1T
#      1e+30            0              0          0          0
#      EPS_1T1          EPS_2T1          SIGMA_RST1      Wpmax_trac_1
#      0                0              0          0
#      sig_trac_2        B_2T            N_2T        SIGMA_2MAXT      C_2T
#      1e+30            0              0          0          0
#      EPS_1T2          EPS_2T2          SIGMA_RST2      Wpmax_trac_2
#      0                0              0          0
#      sig_comp_1        B_1C            N_1C        SIGMA_1MAXC      C_1C
#      1e+30            0              0          0          0
#      EPS_1C1          EPS_2C1          SIGMA_RSC1      Wpmax_comp_1
#      0                0              0          0
#      sig_comp_2        B_2C            N_2C        SIGMA_2MAXC      C_2C
#      1e+30            0              0          0          0
#      EPS_1C2          EPS_2C2          SIGMA_RSC2      Wpmax_comp_2
#      0                0              0          0
#      sig_12           B_12T            N_12T        SIGMA_12MAXT      C_12T
#      .0040            67.0            .29          0          0
#      EPS_1T12          EPS_2T12          SIGMA_RST12      Wpmax_trac_12
#      0.00            0.00            0.00          0
#      GAMMA_INI        GAMMA_MAX        Dmax
#      1E31            1E31            .9999
#      Fsmooth          Fcut
#      0                0
#---1---|---2---|---3---|---4---|---5---|---6---|---7---|---8---|---9---|---10---|

```



Yield
Stress

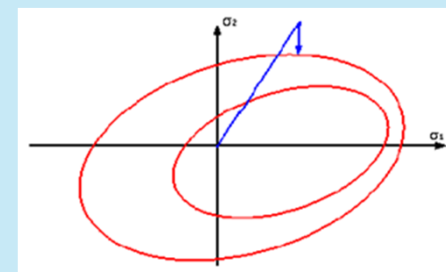
Composite Material Model: Plasticity



```

#---1---|---2---|---3---|---4---|---5---|---6---|---7---|---8---|---9---|---10---|
/MAT/COMPSH/10
Altair composite material
#      Init. dens.      Ref. dens.
#      1.5000E-6        0
#      E11              E22              NU12      IFLAG
#      42              40              .05        1
#      G12              G23              G31        EPSF1
#      3.4              3              3          0
#      ESPT1            EPSM1            EPST2        EPSM2
#      0                0              0          0
#      Wpmax            Wpref            Ioff
#      0                0              5
#      C                EPS              ALPHA        Icc
#      0                0              0          0
#      sig_trac_1        B_1T            N_1T        SIGMA_1MAXT      C_1T
#      1e+30            0              0          0          0
#      EPS_1T1            EPS_2T1        SIGMA_RST1      Wpmax_trac_1
#      0                0              0          0
#      sig_trac_2        B_2T            N_2T        SIGMA_2MAXT      C_2T
#      1e+30            0              0          0          0
#      EPS_1T2            EPS_2T2        SIGMA_RST2      Wpmax_trac_2
#      0                0              0          0
#      sig_comp_1        B_1C            N_1C        SIGMA_1MAXC      C_1C
#      1e+30            0              0          0          0
#      EPS_1C1            EPS_2C1        SIGMA_RSC1      Wpmax_comp_1
#      0                0              0          0
#      sig_comp_2        B_2C            N_2C        SIGMA_2MAXC      C_2C
#      1e+30            0              0          0          0
#      EPS_1C2            EPS_2C2        SIGMA_RSC2      Wpmax_comp_2
#      0                0              0          0
#      sig_12            B_12T          N_12T        SIGMA_12MAXT      C_12T
#      .0040            67.0            .29          0          0
#      EPS_1T12          EPS_2T12        SIGMA_RST12      Wpmax_trac_12
#      0.00            0.00            0.00          0
#      GAMMA_INI          GAMMA_MAX      Dmax
#      1E31            1E31            .9999
#      Fsmooth            Fcut
#      0                0
#---1---|---2---|---3---|---4---|---5---|---6---|---7---|---8---|---9---|---10---|

```



Plasticity,
Strain
Rates

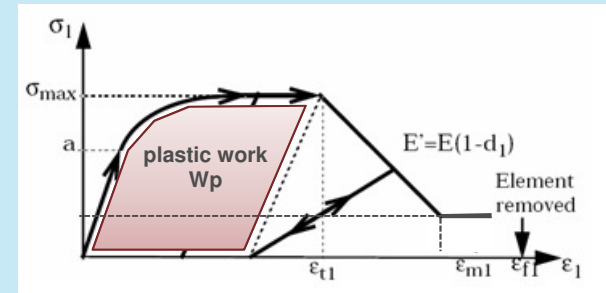
Composite Material Model: Damage and Failure



```

#---1---|---2---|---3---|---4---|---5---|---6---|---7---|---8---|---9---|---10---|
/MAT/COMPSH/10
Altair composite material
#      Init. dens.      Ref. dens.
#      1.5000E-6      0
#      E11      E22      NU12      IFLAG
#      42      40      .05      1
#      G12      G23      G31
#      3.4      3      3
#      ESPT1      EPSM1      EPST2
#      0      0      0
#      Wpmax      Wpref      Ioff
#      0      0      5
#      C      EPS      ALPHA
#      0      0      0
#      sig_trac_1      B_1T      N_1T      SIGMA_1MAXT      C_1T
#      1e+30      0      0      0      0
#      EPS_1T1      EPS_2T1      SIGMA_RST1      Wpmax_trac_1
#      0      0      0      0
#      sig_trac_2      B_2T      N_2T      SIGMA_2MAXT      C_2T
#      1e+30      0      0      0      0
#      EPS_1T2      EPS_2T2      SIGMA_RST2      Wpmax_trac_2
#      0      0      0      0
#      sig_comp_1      B_1C      N_1C      SIGMA_1MAXC      C_1C
#      1e+30      0      0      0      0
#      EPS_1C1      EPS_2C1      SIGMA_RSC1      Wpmax_comp_1
#      0      0      0      0
#      sig_comp_2      B_2C      N_2C      SIGMA_2MAXC      C_2C
#      1e+30      0      0      0      0
#      EPS_1C2      EPS_2C2      SIGMA_RSC2      Wpmax_comp_2
#      0      0      0      0
#      sig_12      B_12T      N_12T      SIGMA_12MAXT      C_12T
#      .0040      67.0      .29      0      0
#      EPS_1T12      EPS_2T12      SIGMA_RST12      Wpmax_trac_12
#      0.00      0.00      0.00      0
#      GAMMA_INI      GAMMA_MAX      Dmax
#      1E31      1E31      .9999
#      Fsmooth      Fcut
#      0      0
#---1---|---2---|---3---|---4---|---5---|---6---|---7---|---8---|---9---|---10---|

```



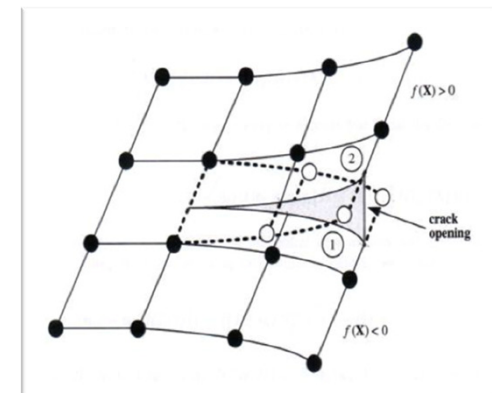
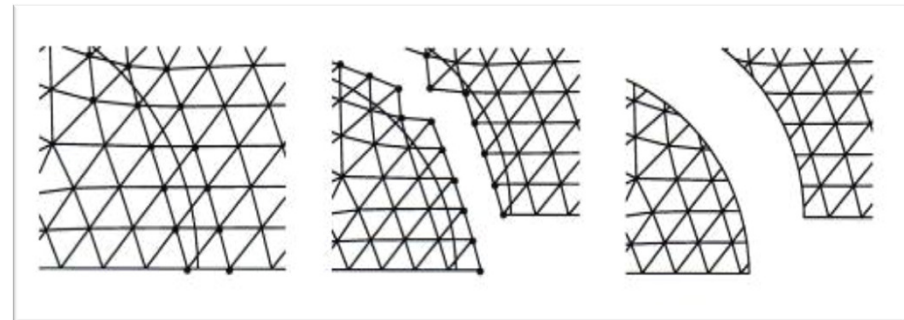
Damage &
Failure

Composite Material Model: Advanced Failure Models

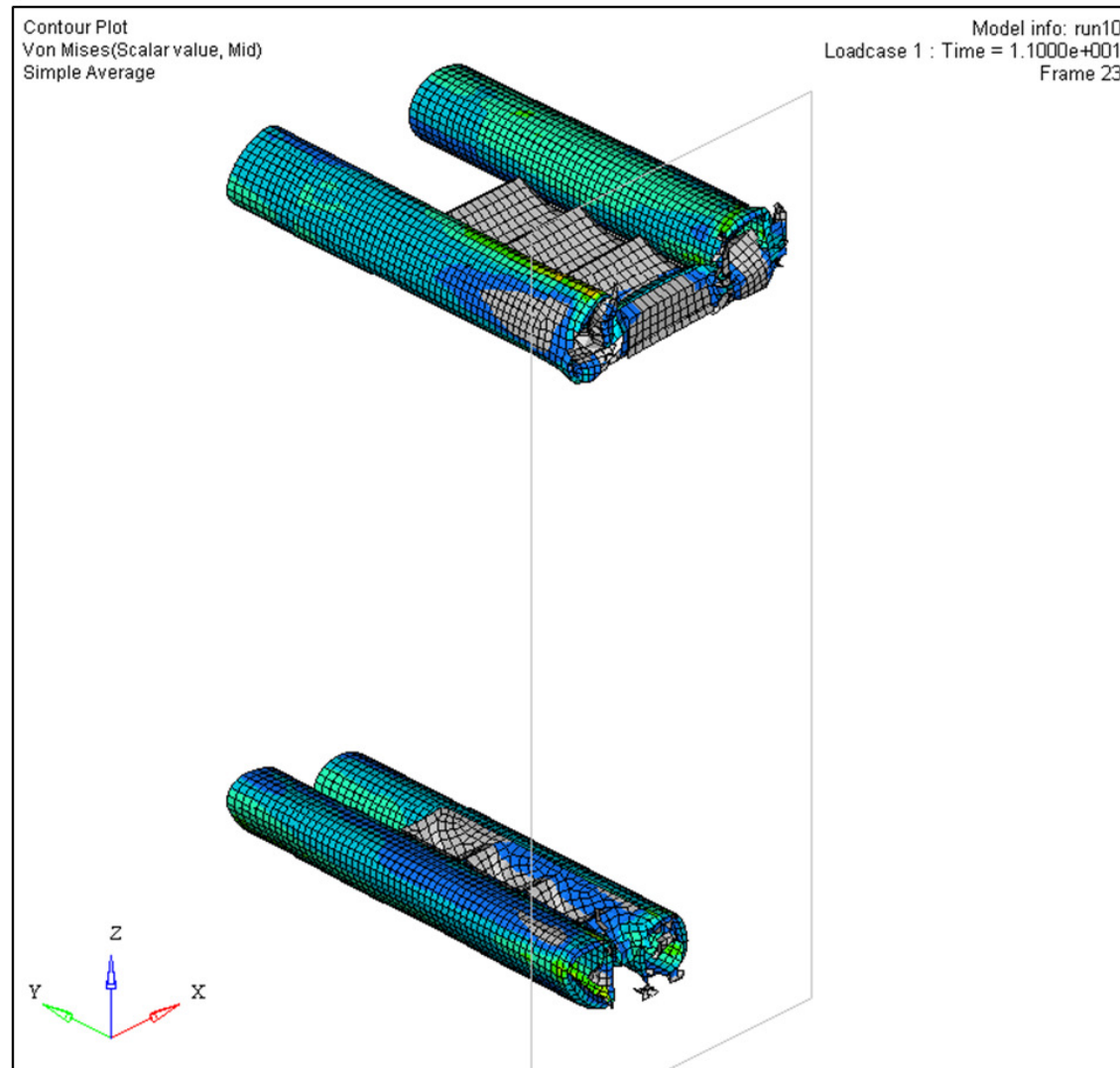


Failure Model Keyword	Type	Description
CHANG	Chang-Chang model	Failure criteria for composites
CONNECT	Failure	Normal and Tangential failure model
ENERGY	Energy isotrop	Specific energy
FLD	Forming limit diagram	Fld
HASHIN	Composite model	Hashin model
JOHNSON	Ductile failure model	Johnson-Cook
LAD_DAMA	Composite delamination	Ladeveze delamination model
NXT	NXT failure criteria	Similar to FLD, but based on stresses
PUCK	Composite model	Puck model
SNCONNECT	Failure	Failure criteria for plastic strain
SPALLING	Ductile + Spalling	Spalling + Johnson-Cook
TAB1	Strain failure model	based on damage accumulation using user-defined functions
TBUTCHER	Tuler-Butcher model	Failure due to fatigue
TENSSTRAIN	Traction	Strain failure
USERi	User failure model	
WIERZBICKI	Ductile material	Bao-Xue-Wierzbicki model
WILKINS	Ductile failure model	Wilkins model
XFEM_FLD	Forming limit diagram	Fld
XFEM_JOHNS	Ductile failure model	Johnson-Cook
XFEM_TBUTC	Ductile (brittle) failure model	Modified Tuler-Butcher model

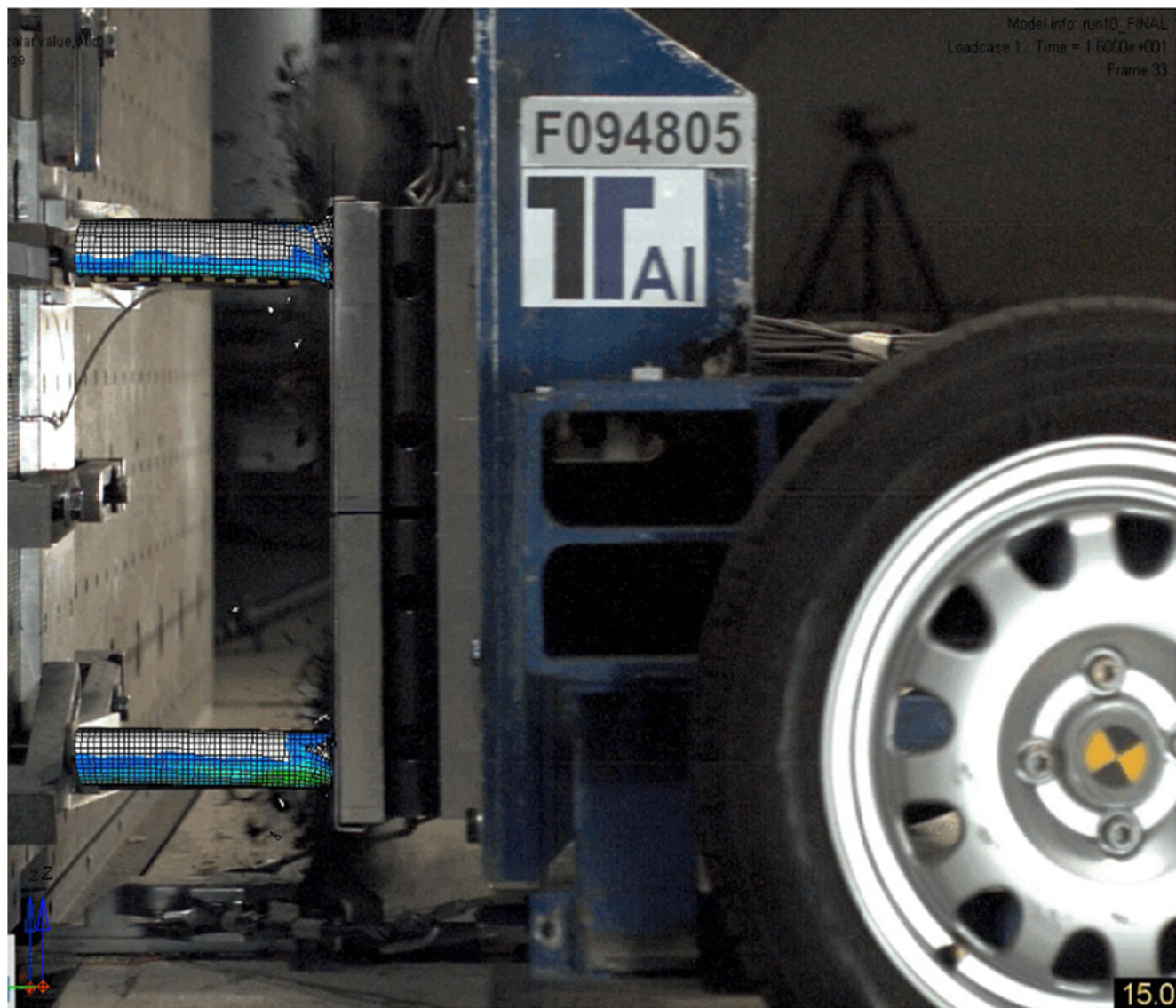
Failure models can be coupled with compatible material laws



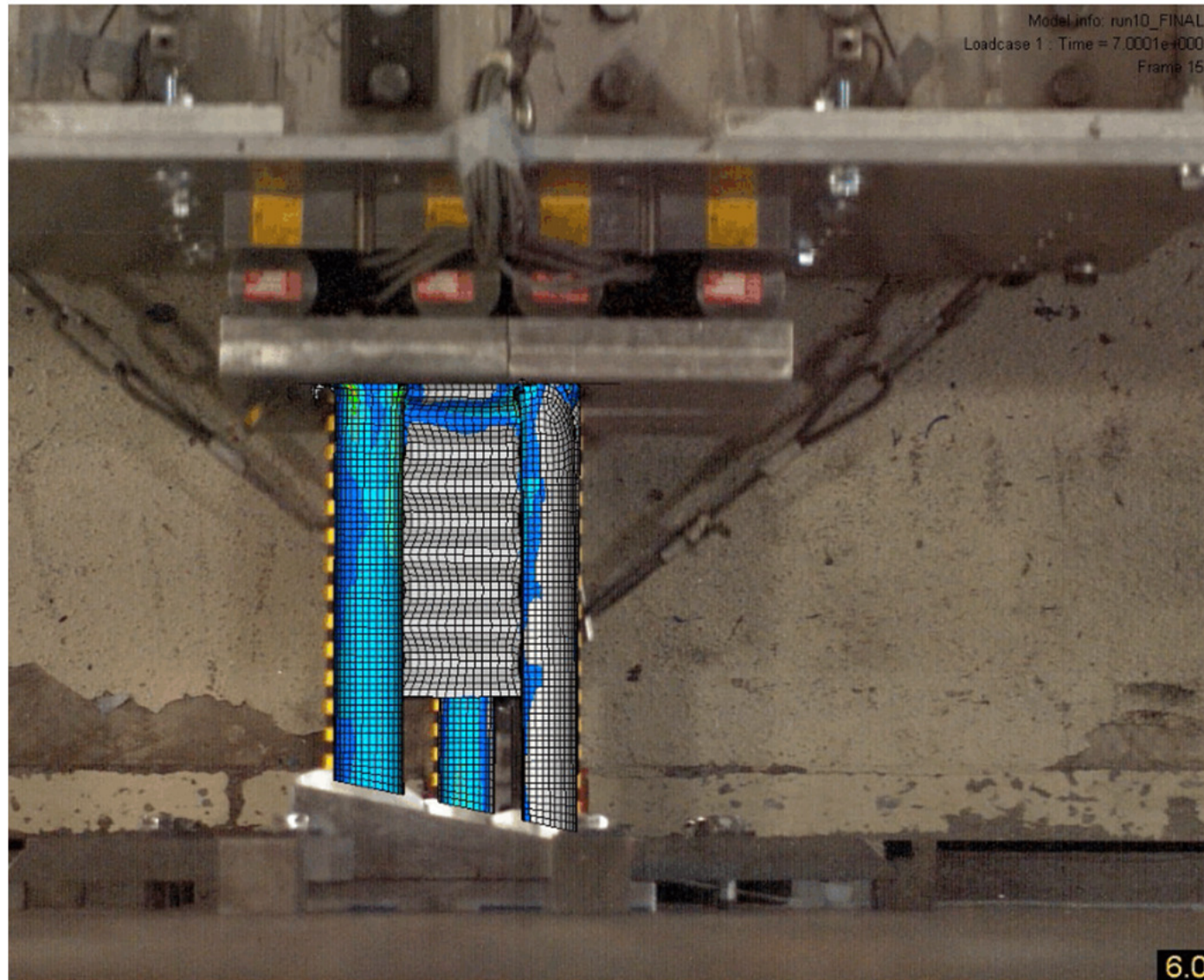
Ergebnisse



Ergebnisse

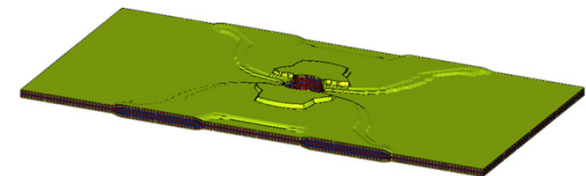


Ergebnisse



Agenda

- Materialmodellierung
 - Validierung einer Crashberechnung
- **Auslegung von Composite Strukturen**
 - **Beschnitt: Optimierung Size/Shape/Shuffling**
 - **Fertigung: Drapiersimulation**



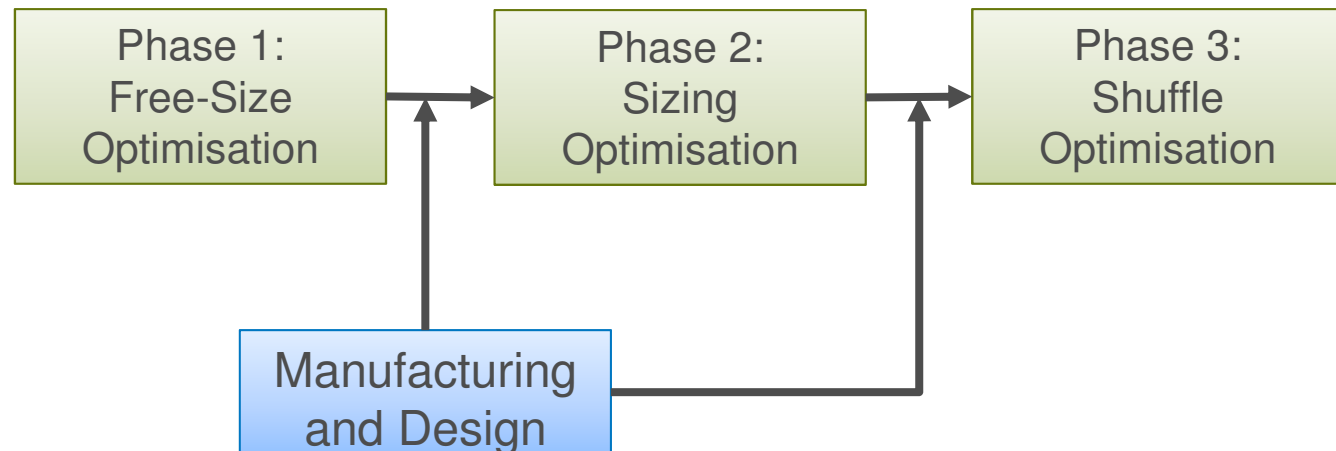
CAE Process Chain

What is known ?

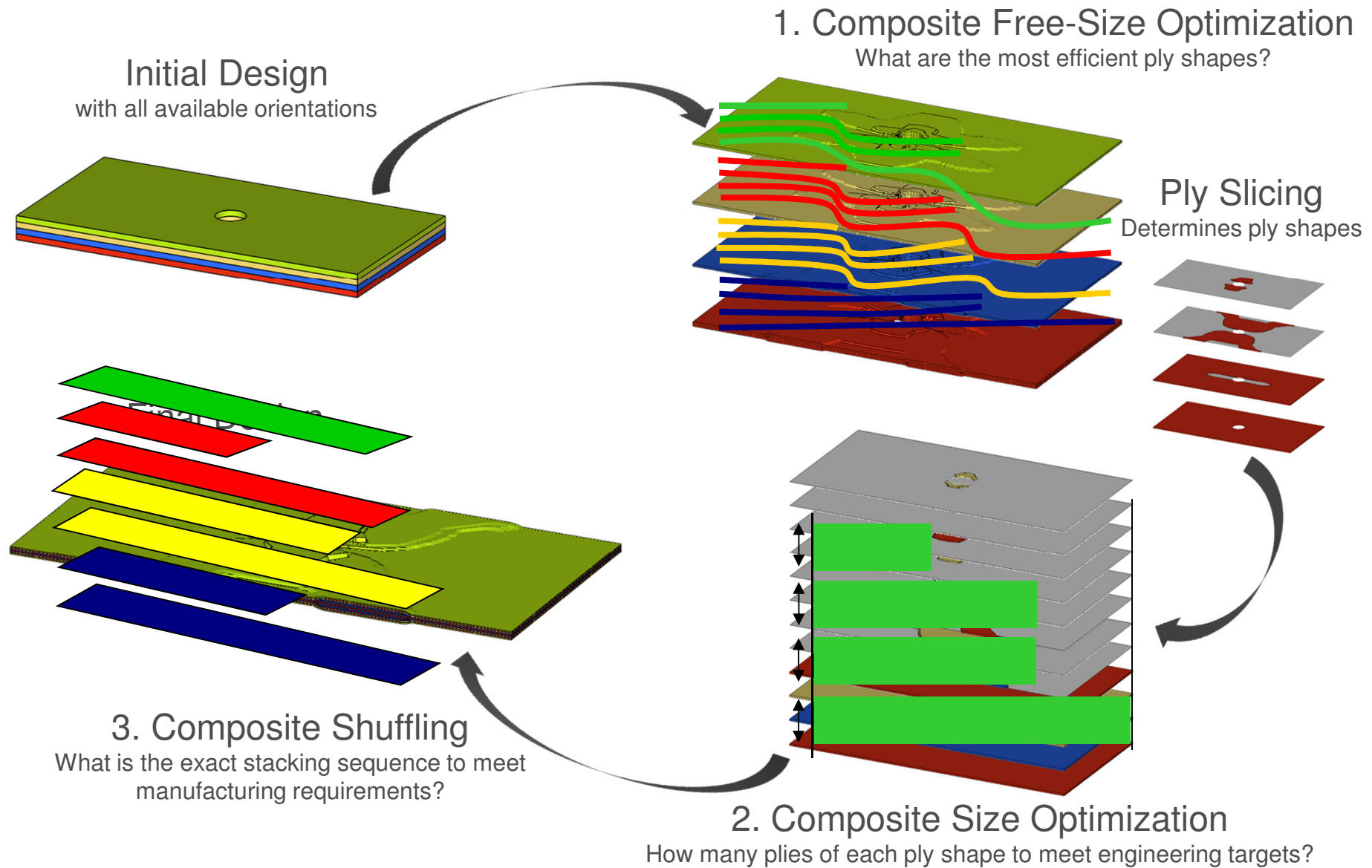
- Manufacturing process
- Design idea
- Available material (fabric, UD-layers, fiber/matrix)
- Possible fiber orientations

What is not known ?

- Distribution of material and fiber orientation
- Final lay-up of the laminate



OptiStruct: Composite Design Process



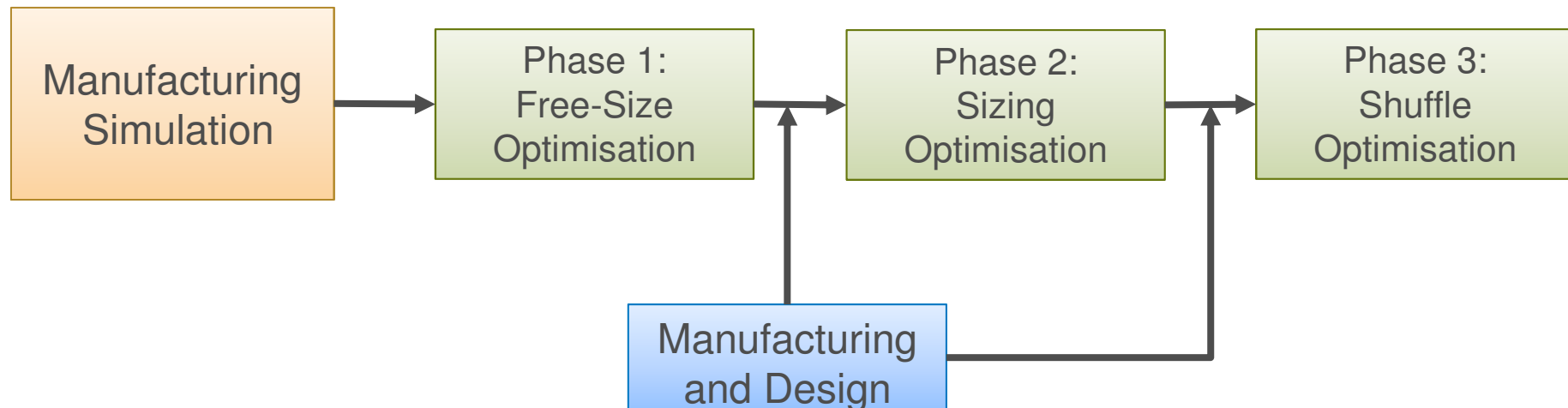
CAE Process Chain

What is known ?

- Manufacturing process
- Design idea
- Available material (fabric, UD-layers, fiber/matrix)
- Possible fiber orientations

What is not known ?

- Distribution of material and fiber orientation
- Final lay-up of the laminate



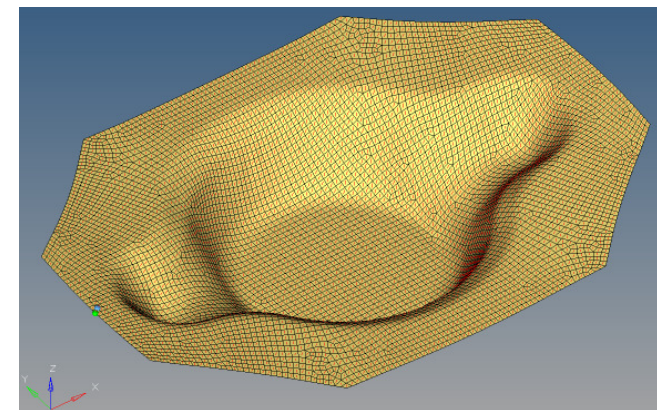
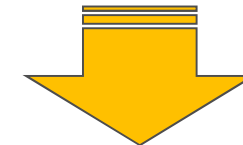
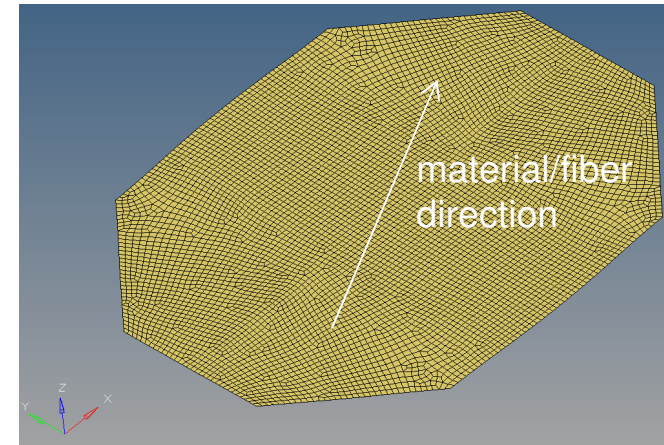
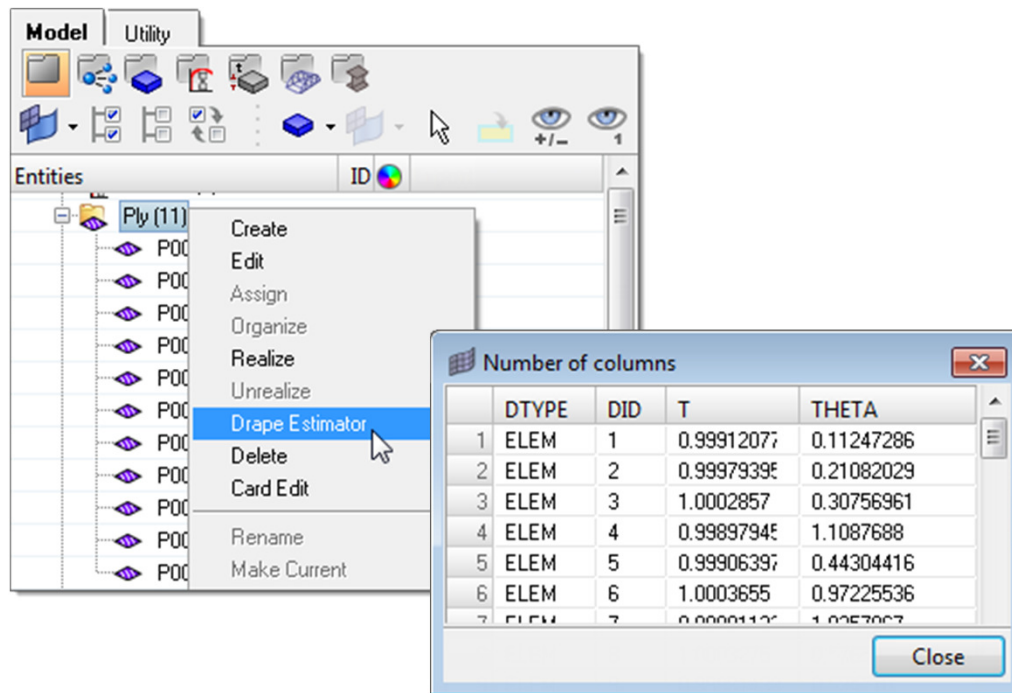
In addition, we can answer:

- Is it possible to manufacture my design at all?
- What is the influence of the manufacturing on the fiber orientation?
- Do I have local thickening or thinning?

HyperMesh: Drape Estimator for Composite Fibers



- Calculate draping angles and thickness variation
- Geometry based, very fast (seconds)
- Good estimation for deviation of the fiber orientation, but feedback on manufacturability only indirect



Radioss: Draping Simulation

Sandwich method

1 Component
1 Material
1 Property

Sandwich material law

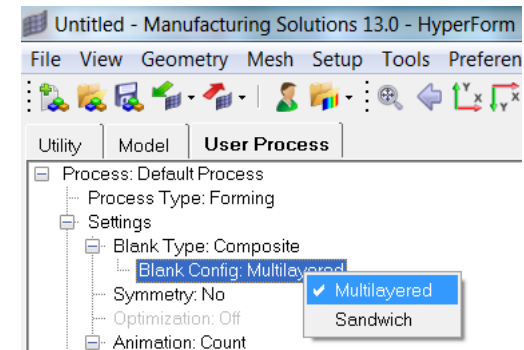
- All plies defined in the same property

Independant layers method

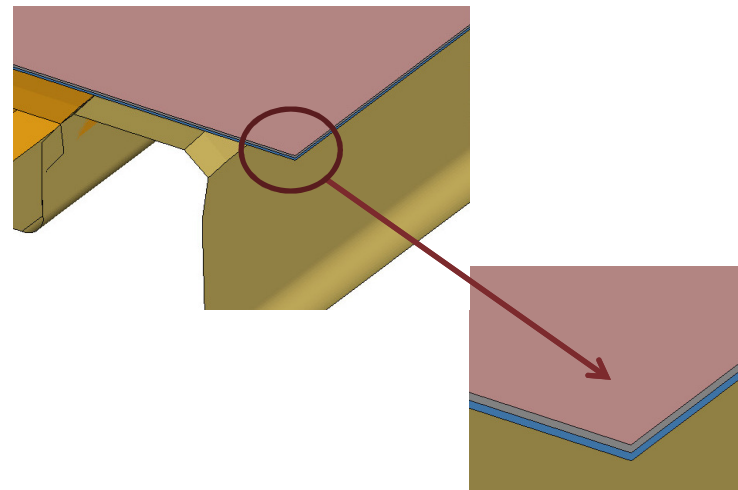
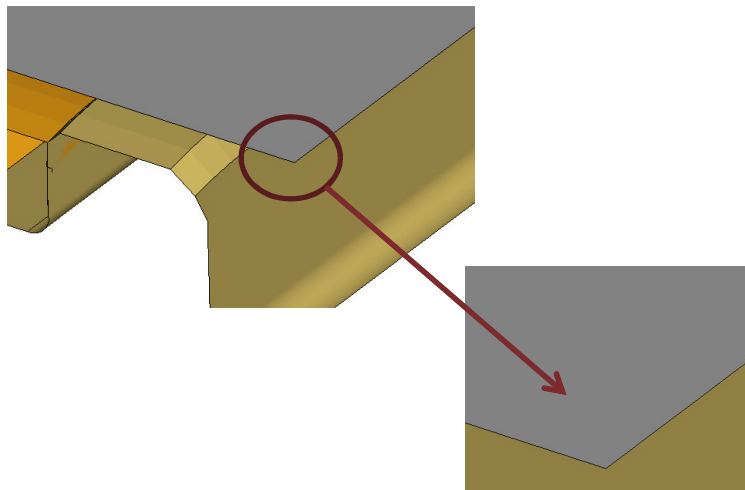
1 Component for each Ply
1 Material for each Ply
1 Property for each Ply

Fabric material law

- All layers defined independently
- Contact interface between the layers



Setup in HyperForm
user process

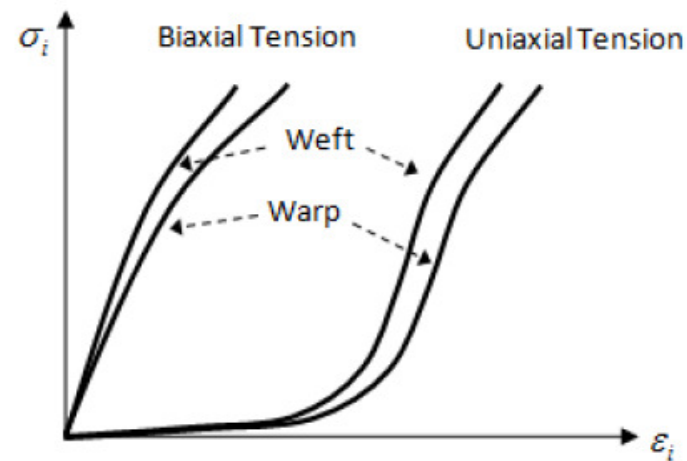
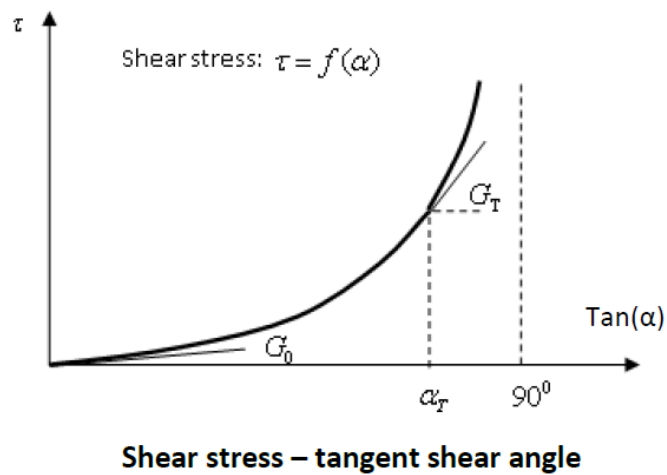
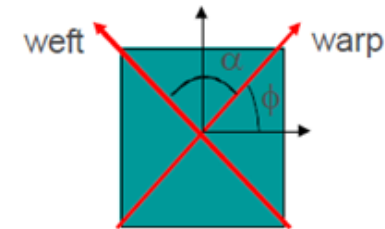


Radioss: Fabric Material Characterization



Capture complete physical ply behavior

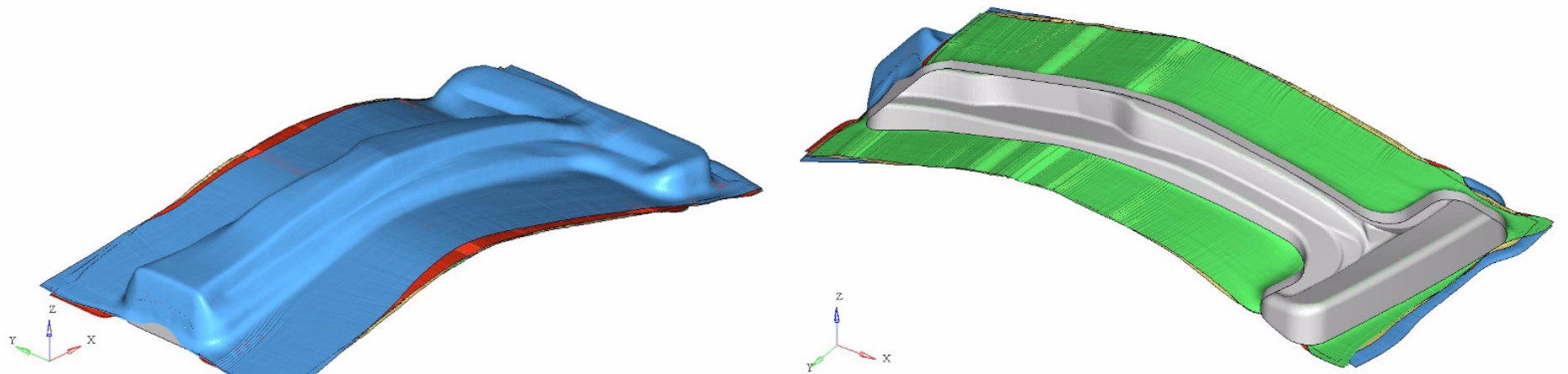
- Stress-strain curve in warp and weft direction
- Shear stress as a function of shear angle
- ...



Radioss: Independent Layers Simulation

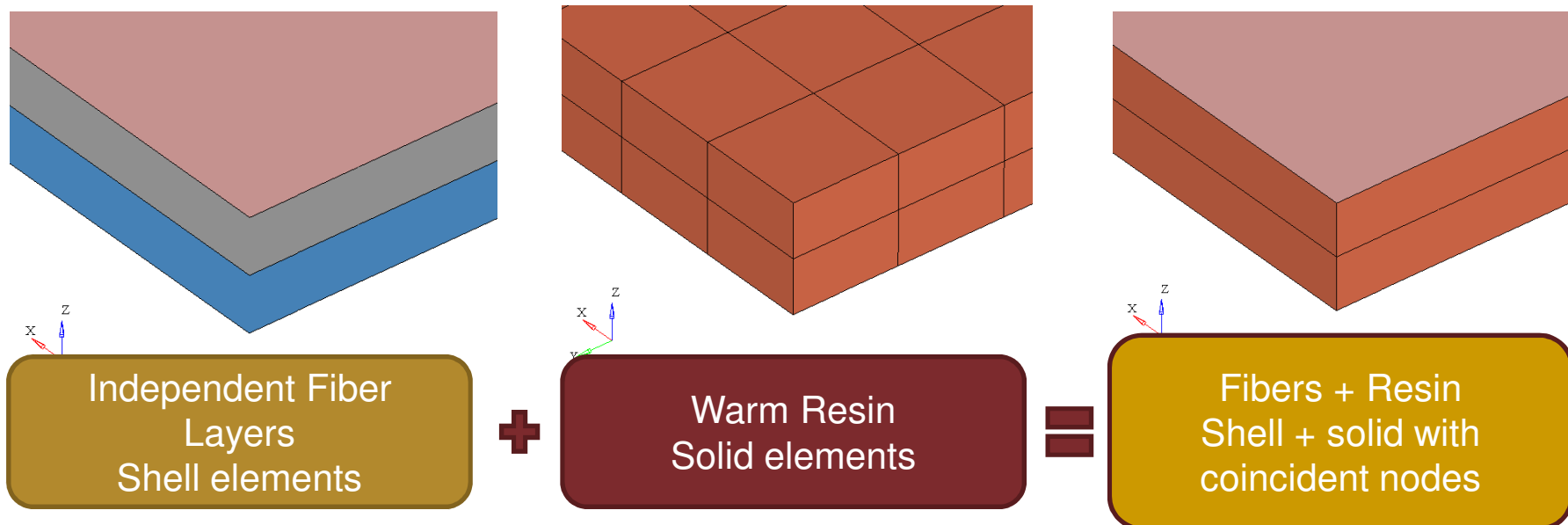
Sliding

- Sliding effect between layers can be modeled by contact interface
- Each layer is free to slide over other layers
- In this example, fiber orientations are 0, +45, 90 and -45° regarding X axis
- The behavior during stamping is different for each layer



Radioss: Independent Layers with Resin

Independant layers + resin method



Connect material is used to model resin :

- A plasticity domain is reached after a small yield stress value
- Strain rate dependancy available for user defined curves

Summary: CAE Process Chain with HyperWorks

